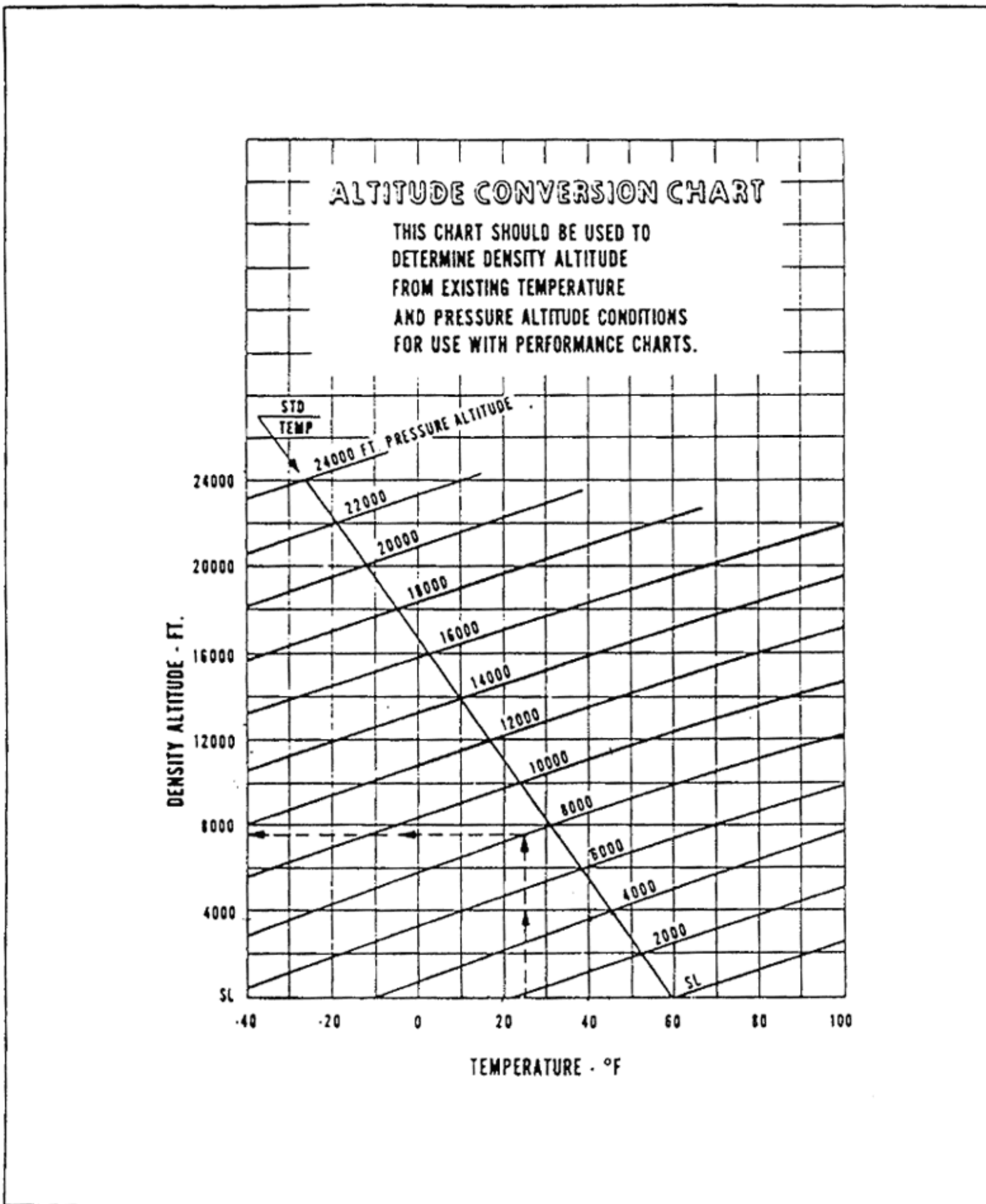


PERFORMANCE CHARTS

Altitude Conversion Chart	9-1
Take-off Ground Run Distance (Flap Setting 0°)	9-2
Take-off Ground Run Distance (Flap Setting 25°)	9-3
Take-off Distance Over 50 Ft (Flap Setting 25°)	9-4
Take-off Distance Over 50 Ft (Flap Setting 0°)	9-5
Accelerate-Stop Distance	9-6
Multi-Engine Climb Performance	9-7
Single Engine Climb Performance	9-8
Cruise Performance - True Airspeed	9-9
Cruise Performance - Range	9-10
Stalling Speed vs Weight	9-11
Stalling Speed vs Angle of Bank	9-12
Landing Ground Run Distance	9-13
Landing Distance Over 50 Ft	9-14
Power Setting Table	9-16

WARNING

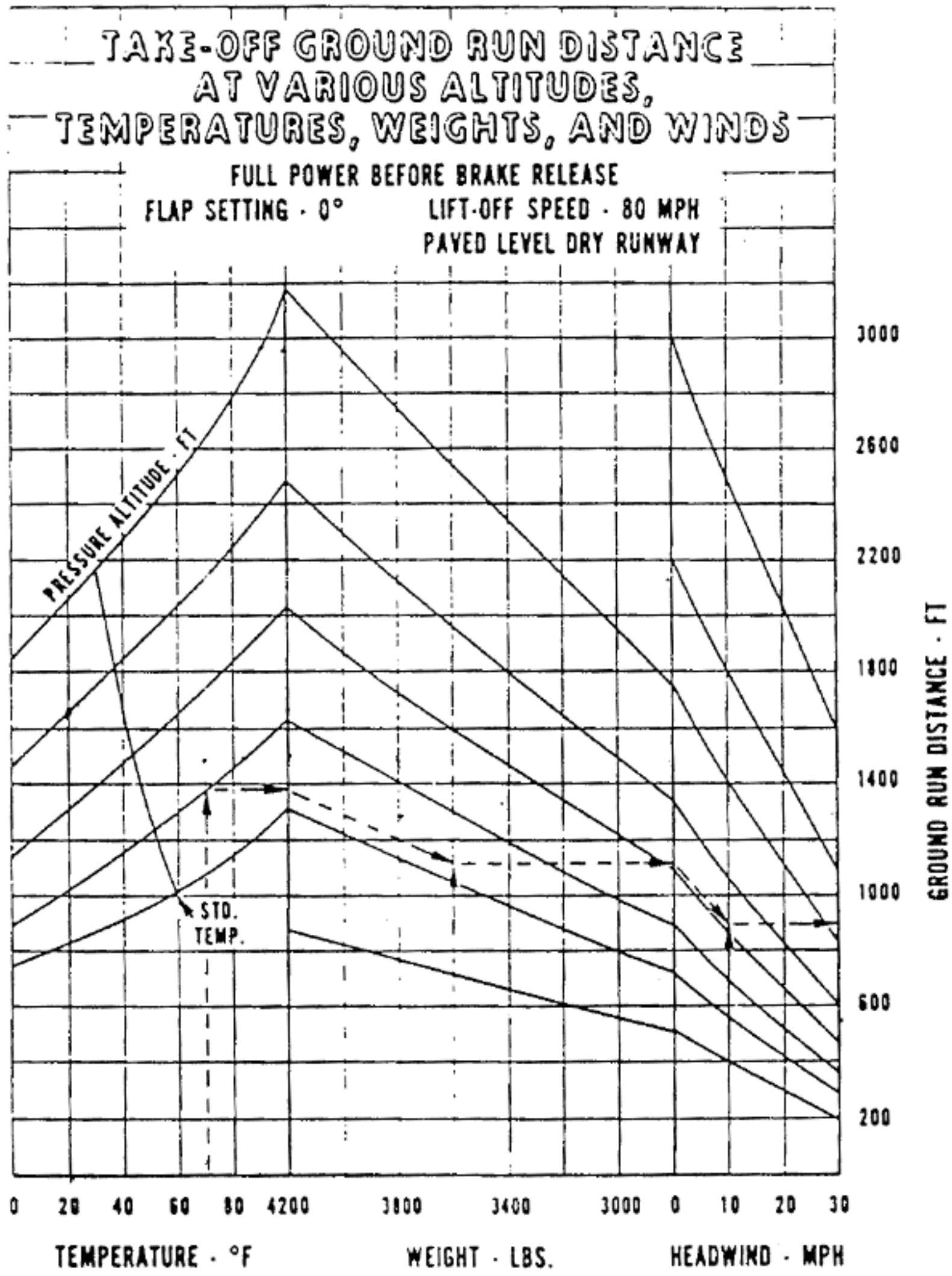
Performance information derived by extrapolation beyond the limits shown on the charts should not be used for flight planning purposes.



Example: Temp. 25° F
Density Alt. 7500 Ft.

Press. Alt. 8000 Ft.

761577

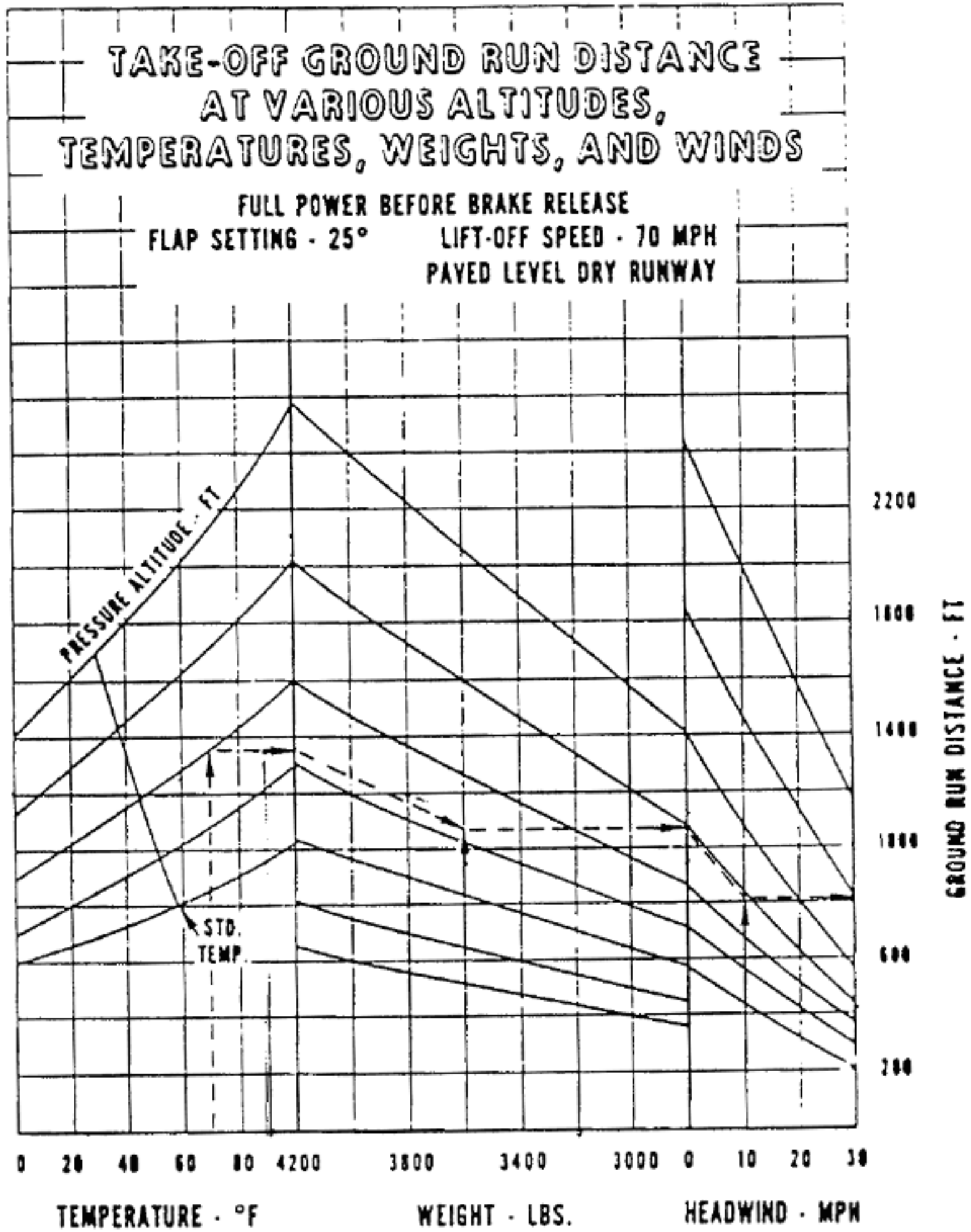


Example: Temp. 70° F
Press. Alt. 2000 Ft.

Wt. 3600 lbs.
Hd. wind 10 MPH

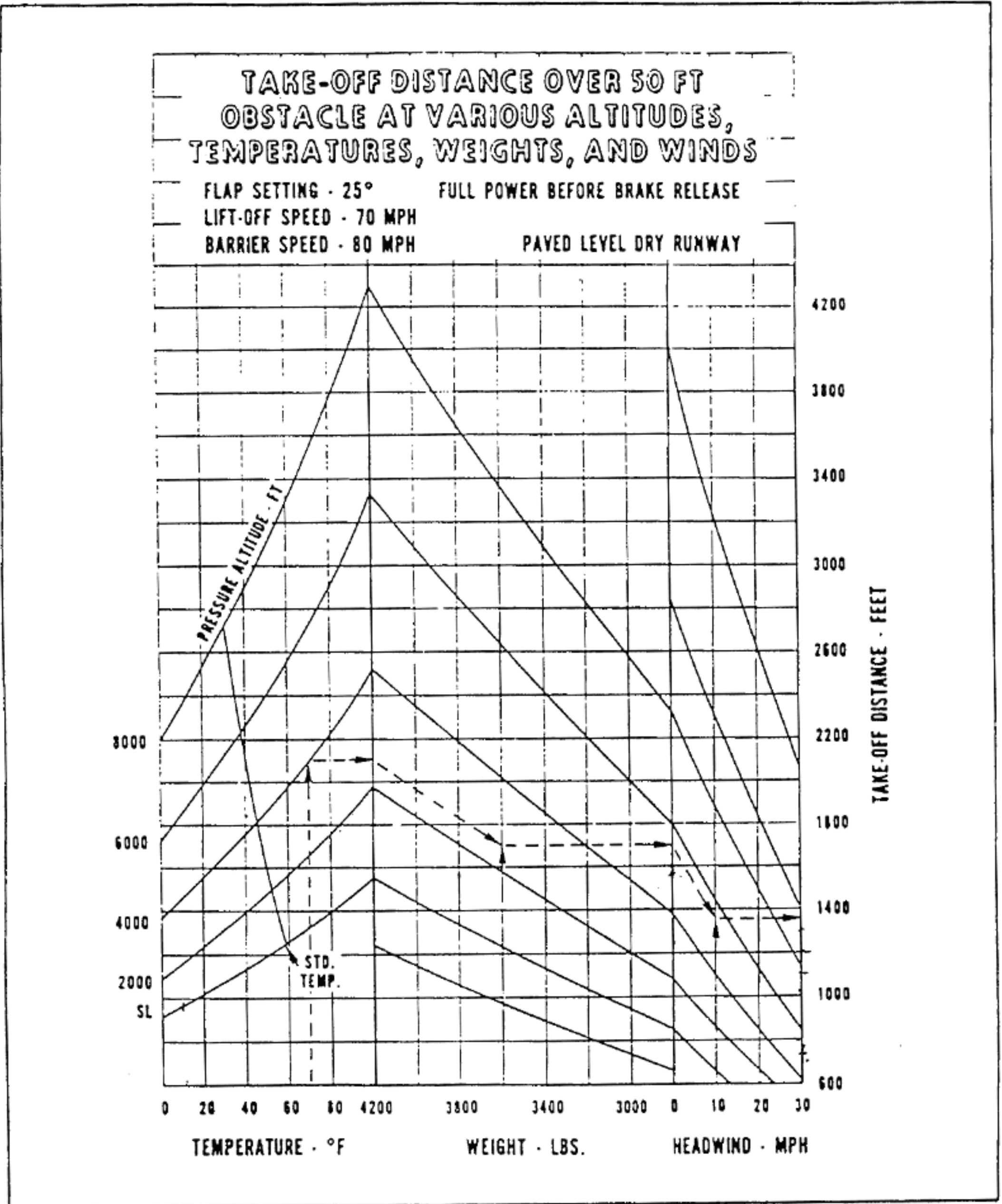
Ground run 900 ft

761 577



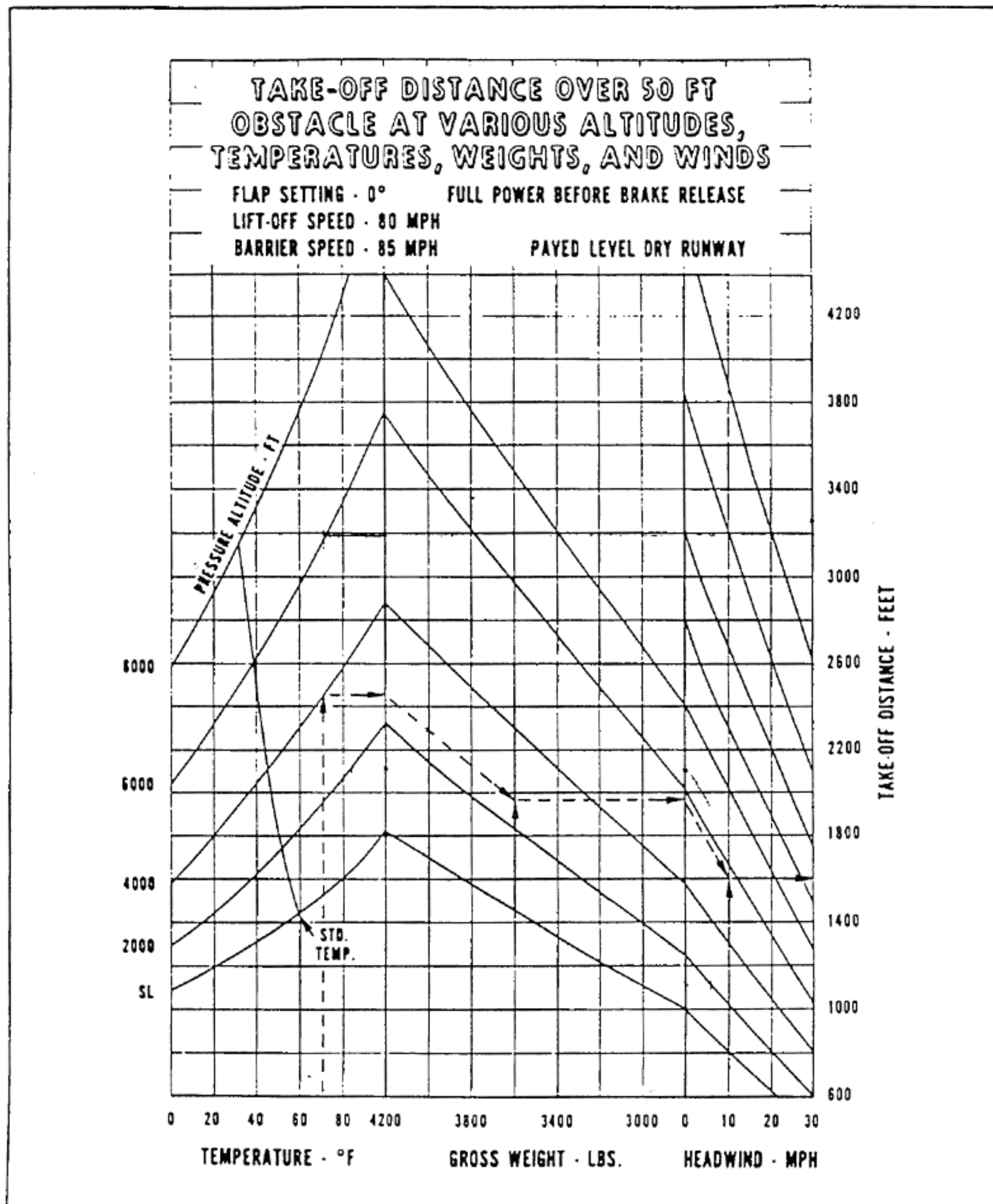
Example: Temp. 70° F Wt. 3600 lbs Ground Run 810 ft
 Press. Alt. 4000 ft Hd. wind 10 MPH

761 577



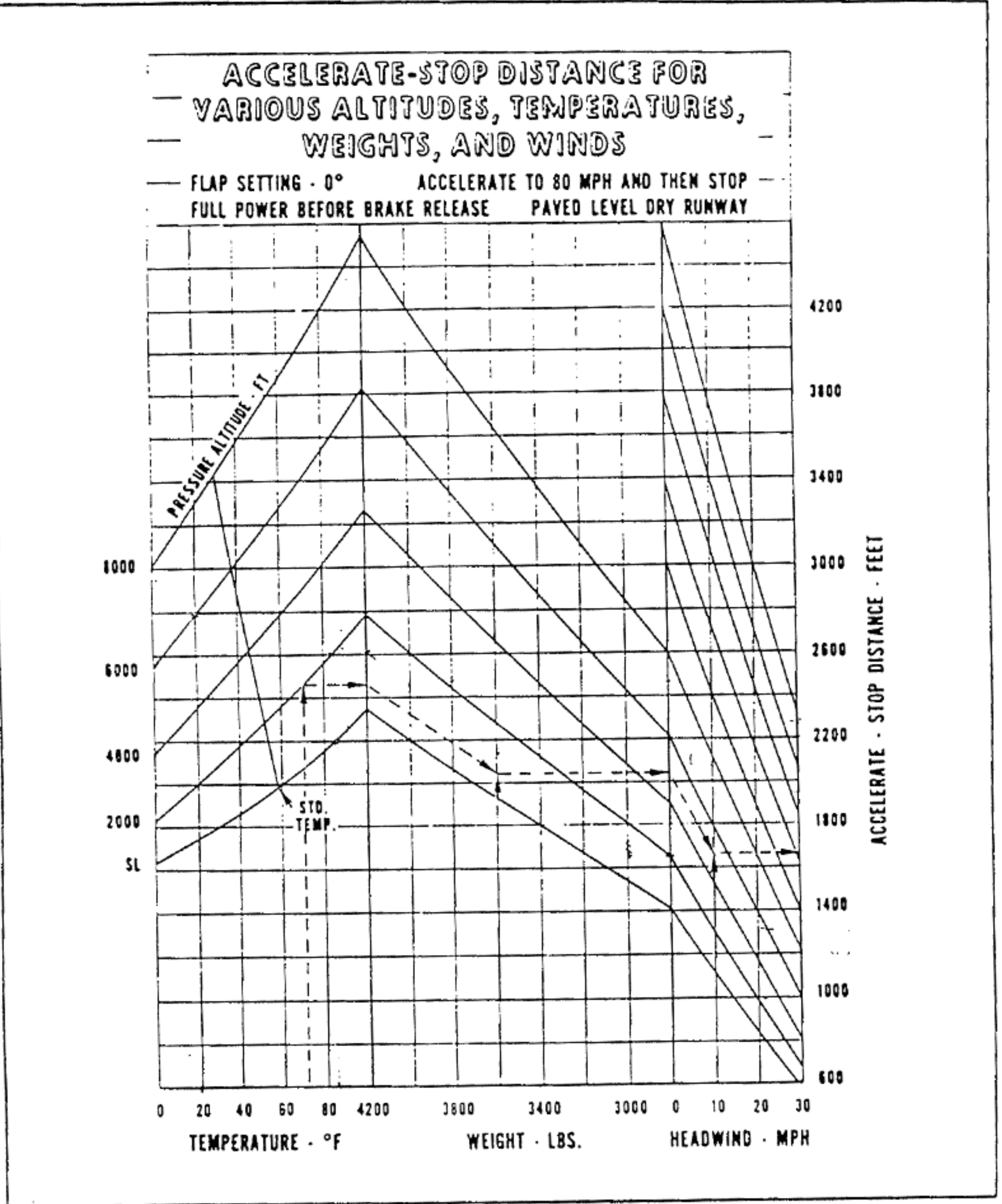
Example: Temp. 70° F Wt. 3600 lbs T. O. Dist. 1350 ft
 Press. Alt. 4000 ft Hd. wind 10 MPH

761 577



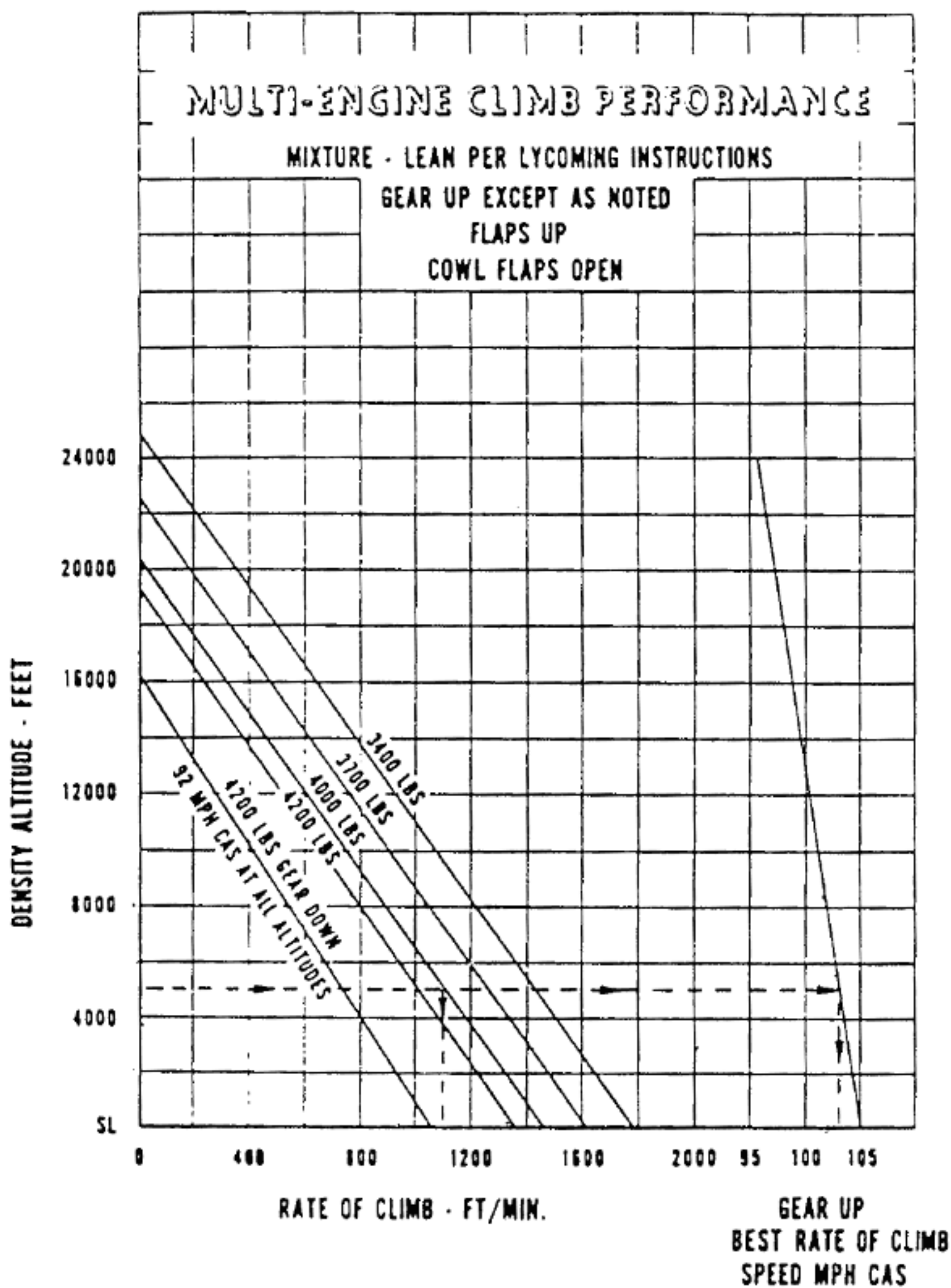
Example: Temp. 70° F Wt. 3600 lbs T. O. Dist. 1600 ft
 Press. Alt. 4000 ft Hd. wind 10 MPH

761 577



Example: Temp. 70° F Wt. 3600 lbs Accel. - Stop Dist. 1650 ft
 Press. Alt. 2000 ft Hd. wind 10 MPH

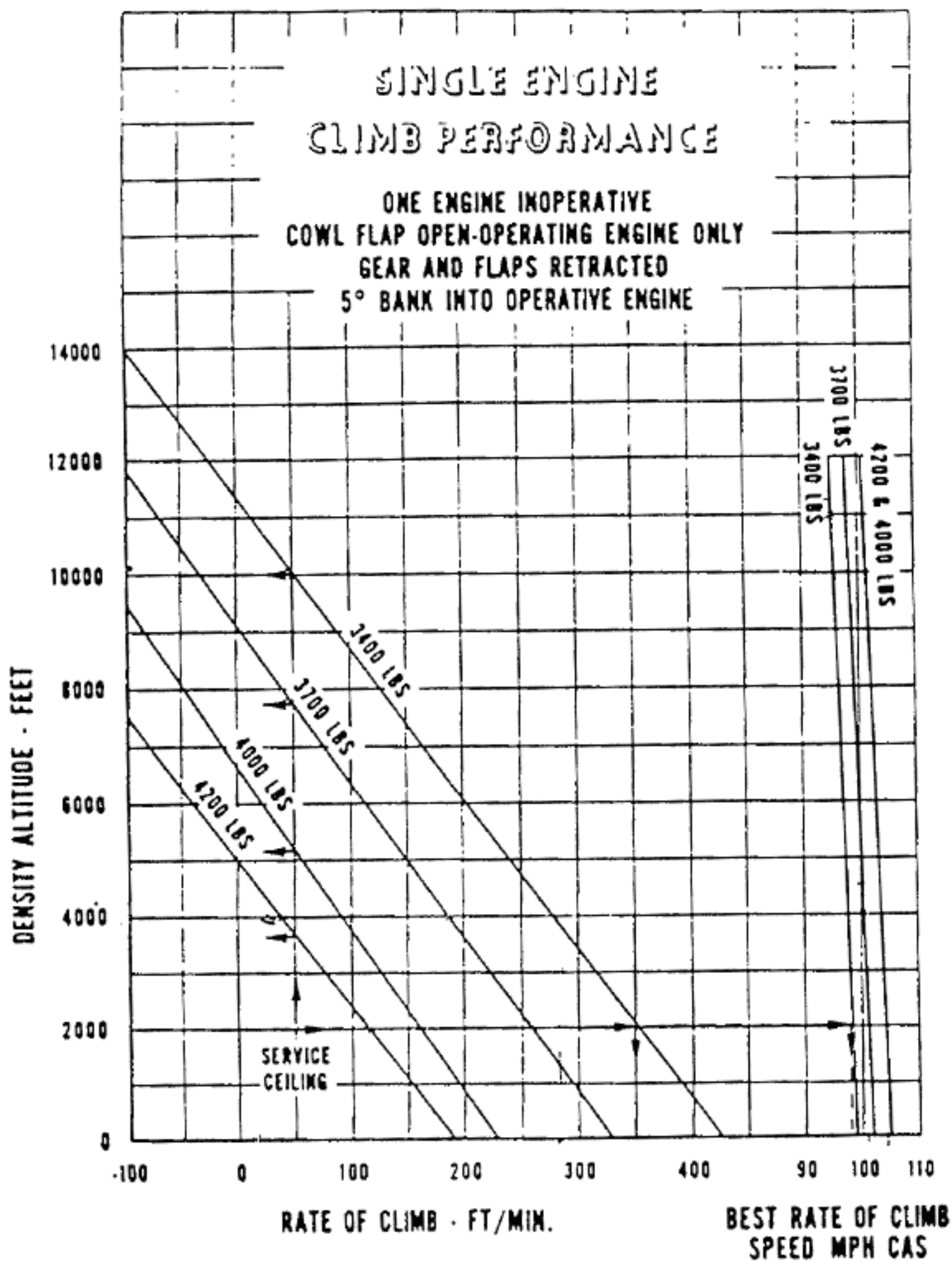
761 577



*Gear down best rate of climb speed is 92 MPH CAS at all altitudes.

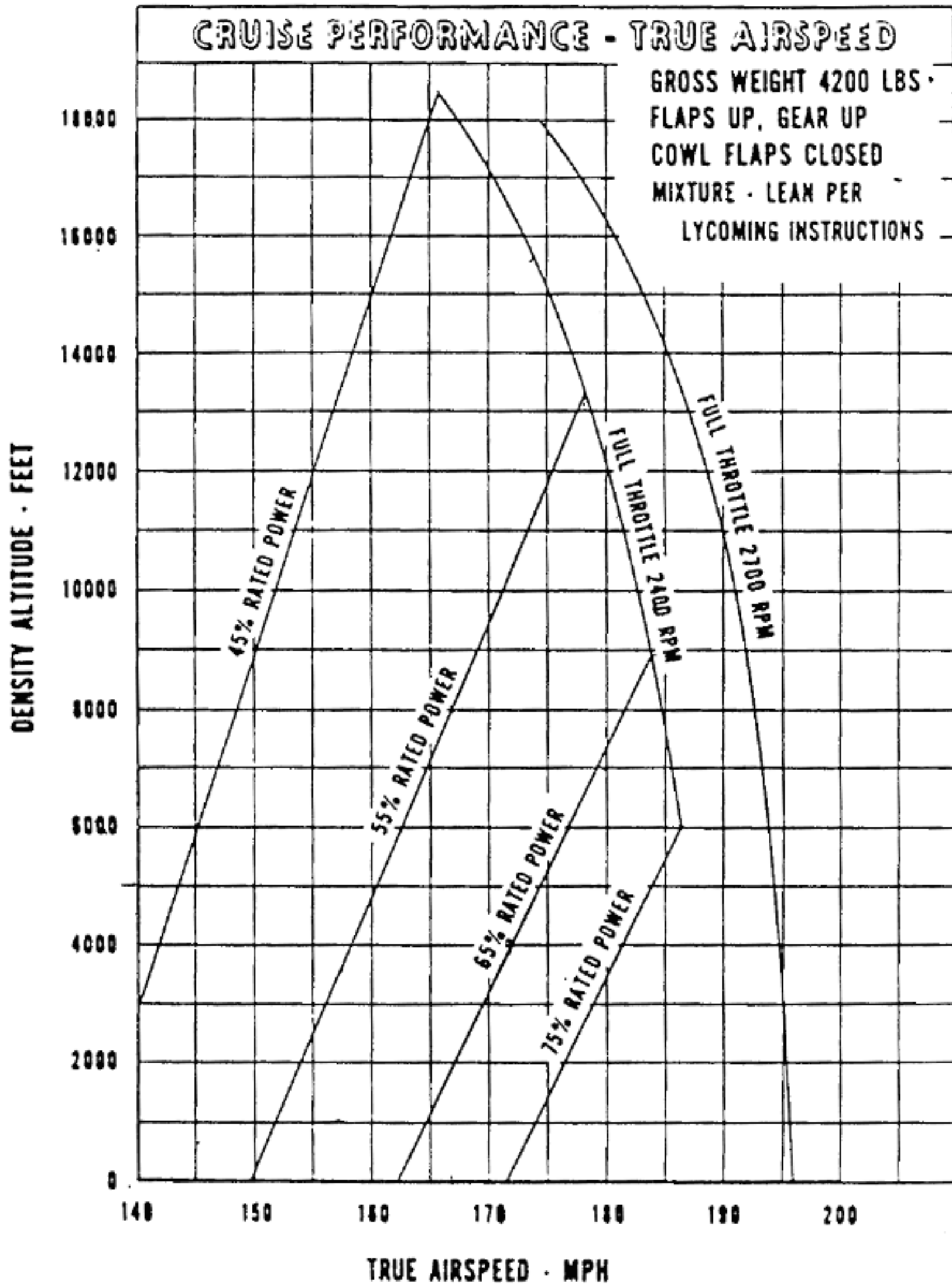
Example: Wt. 4000 lbs Rate of Climb 1100 ft/min
 Den. Alt. 5000 ft Best R/C Speed 103 MPH
 (Gear Up)

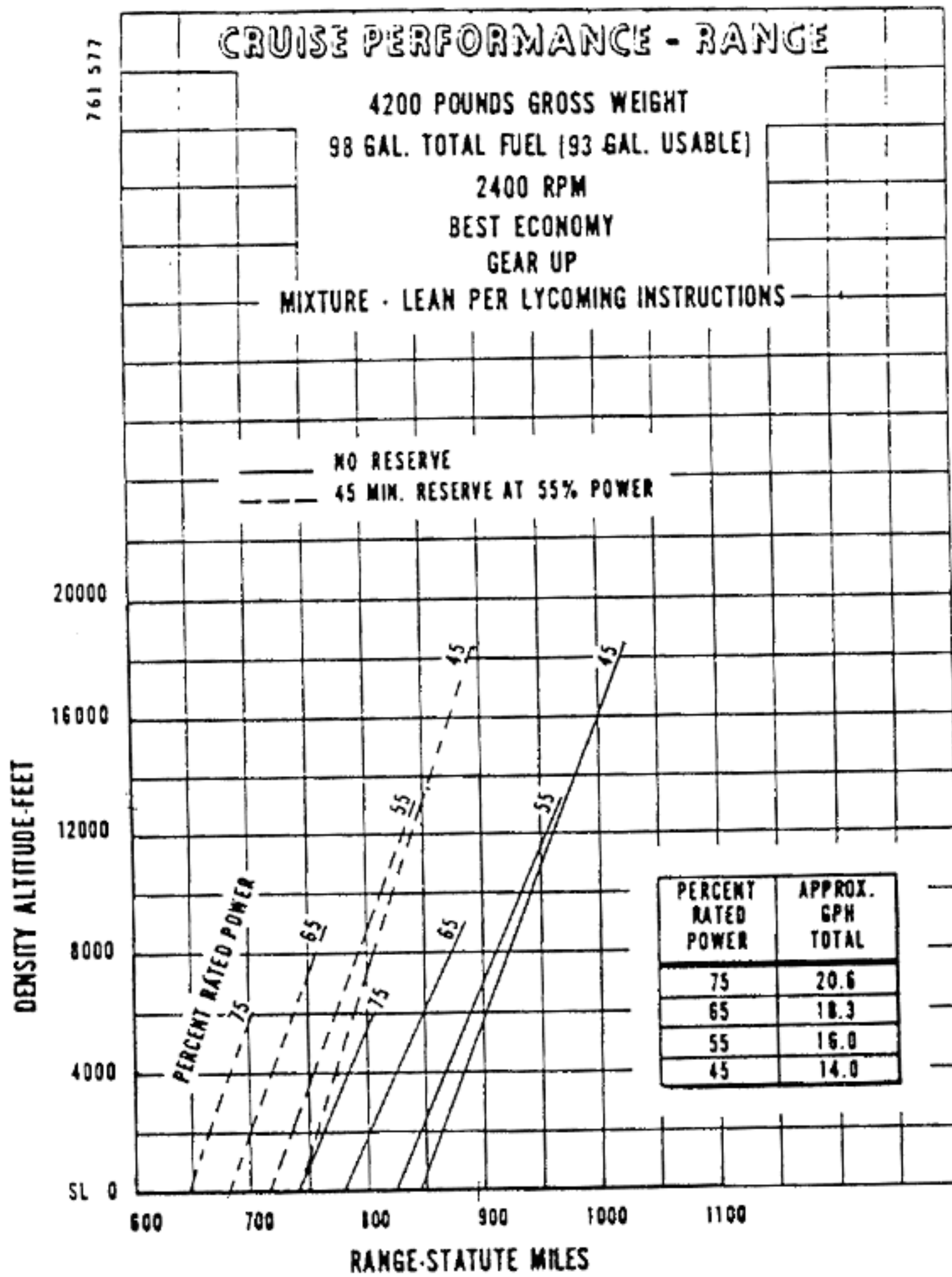
761 577



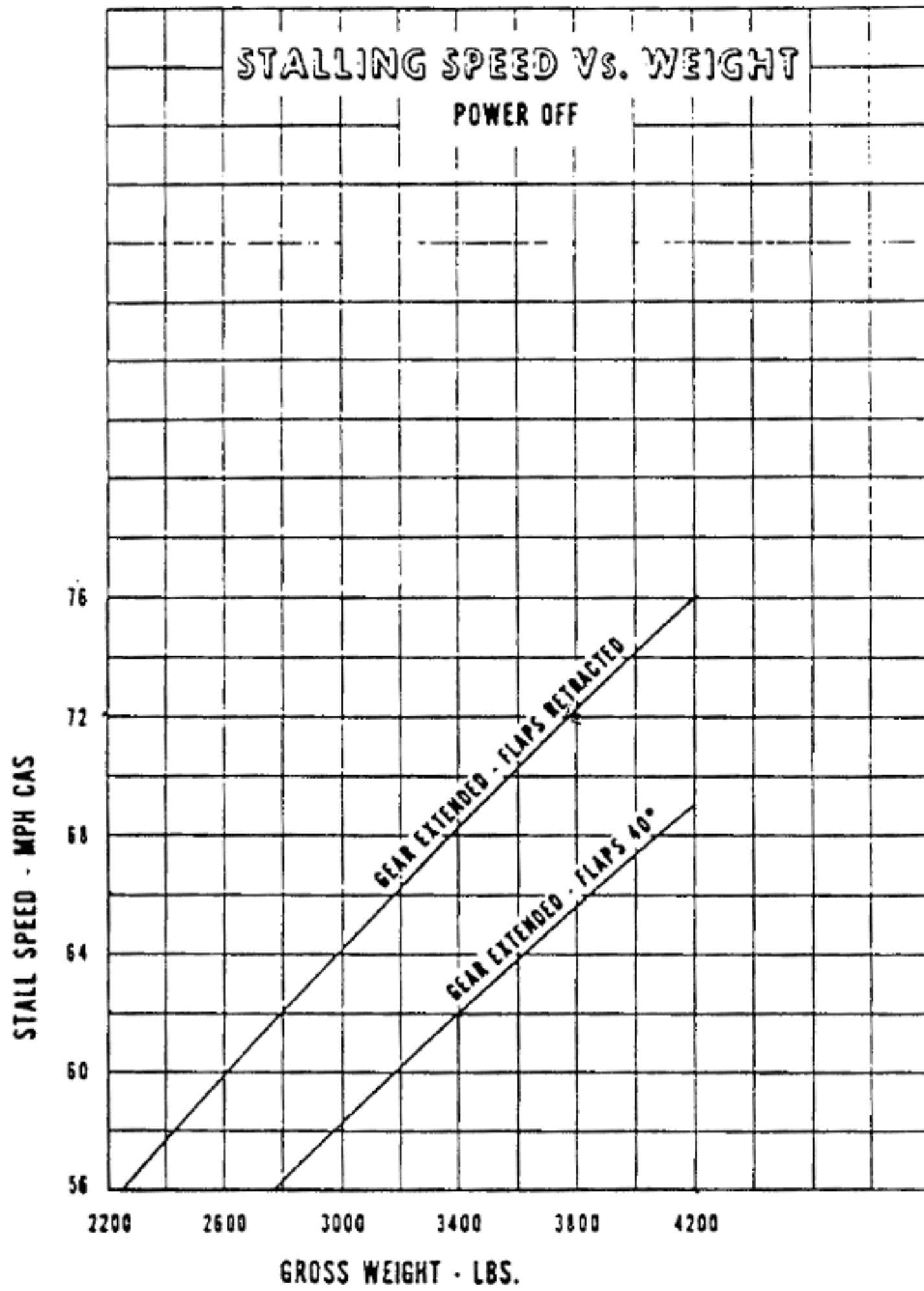
Example: Wt. 3400 lbs Rate of Climb 350 ft/min
 Den. Alt. 2000 ft Best R/C Speed 98 MPH

761 577

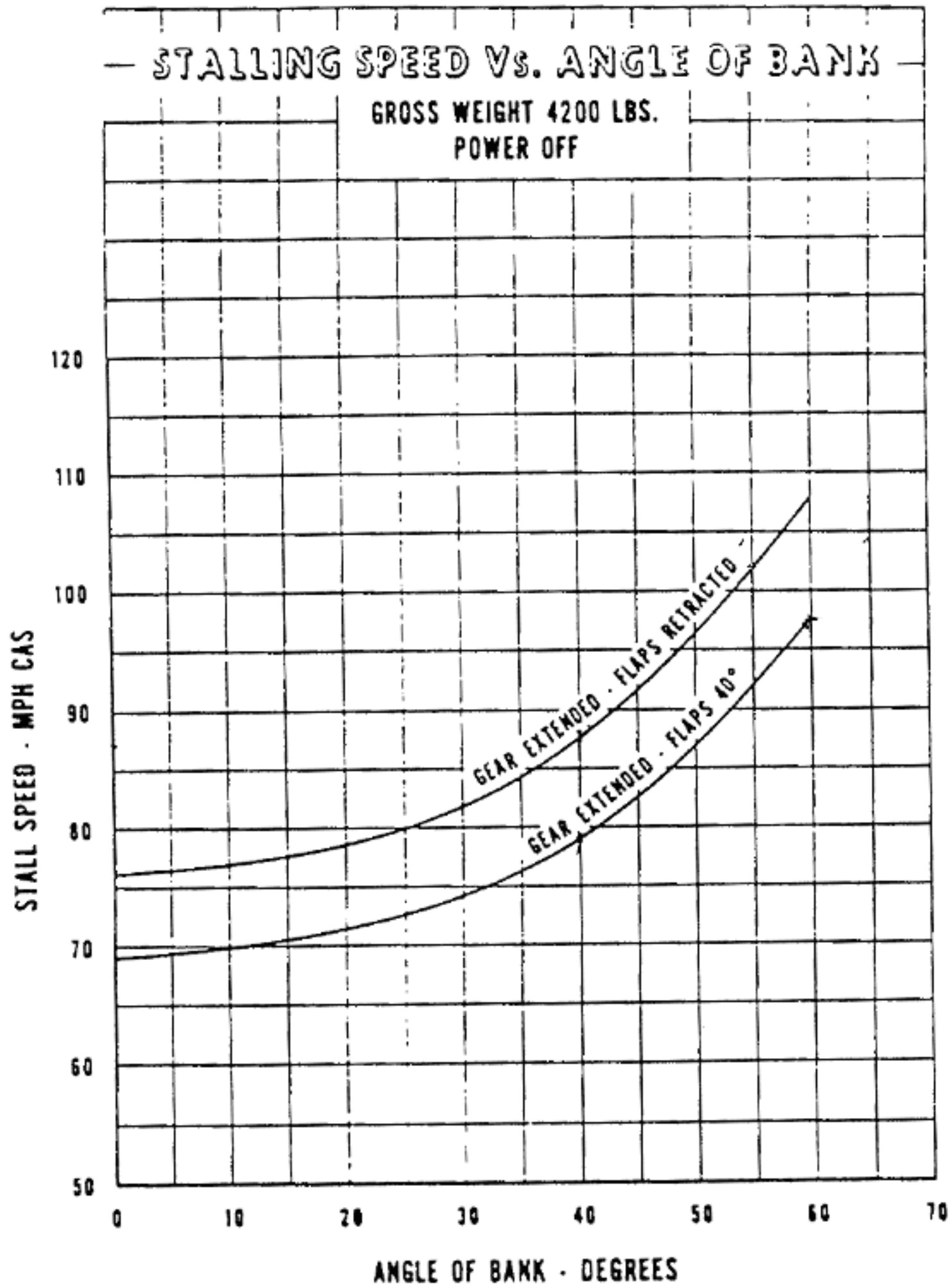




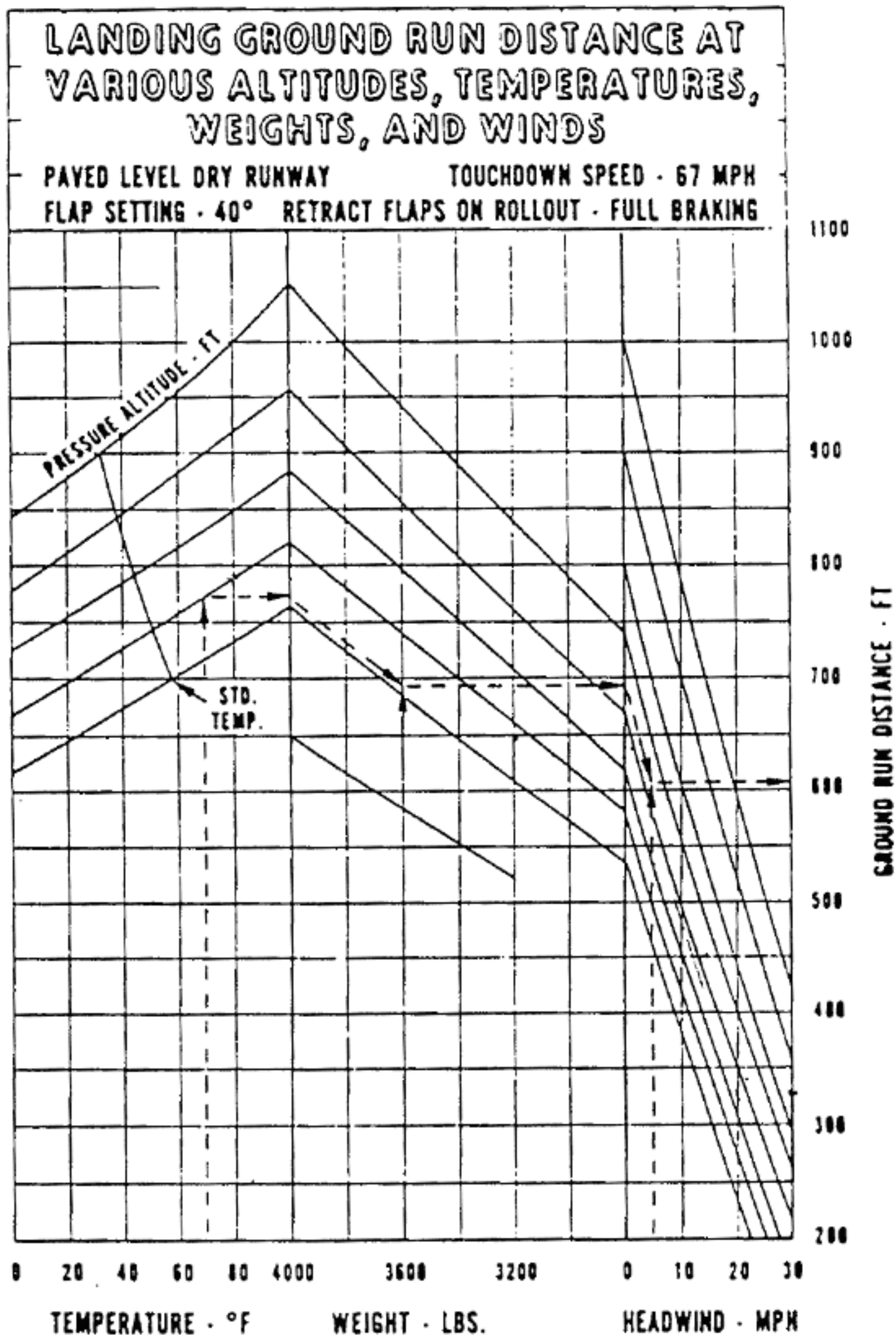
761 677



761 577



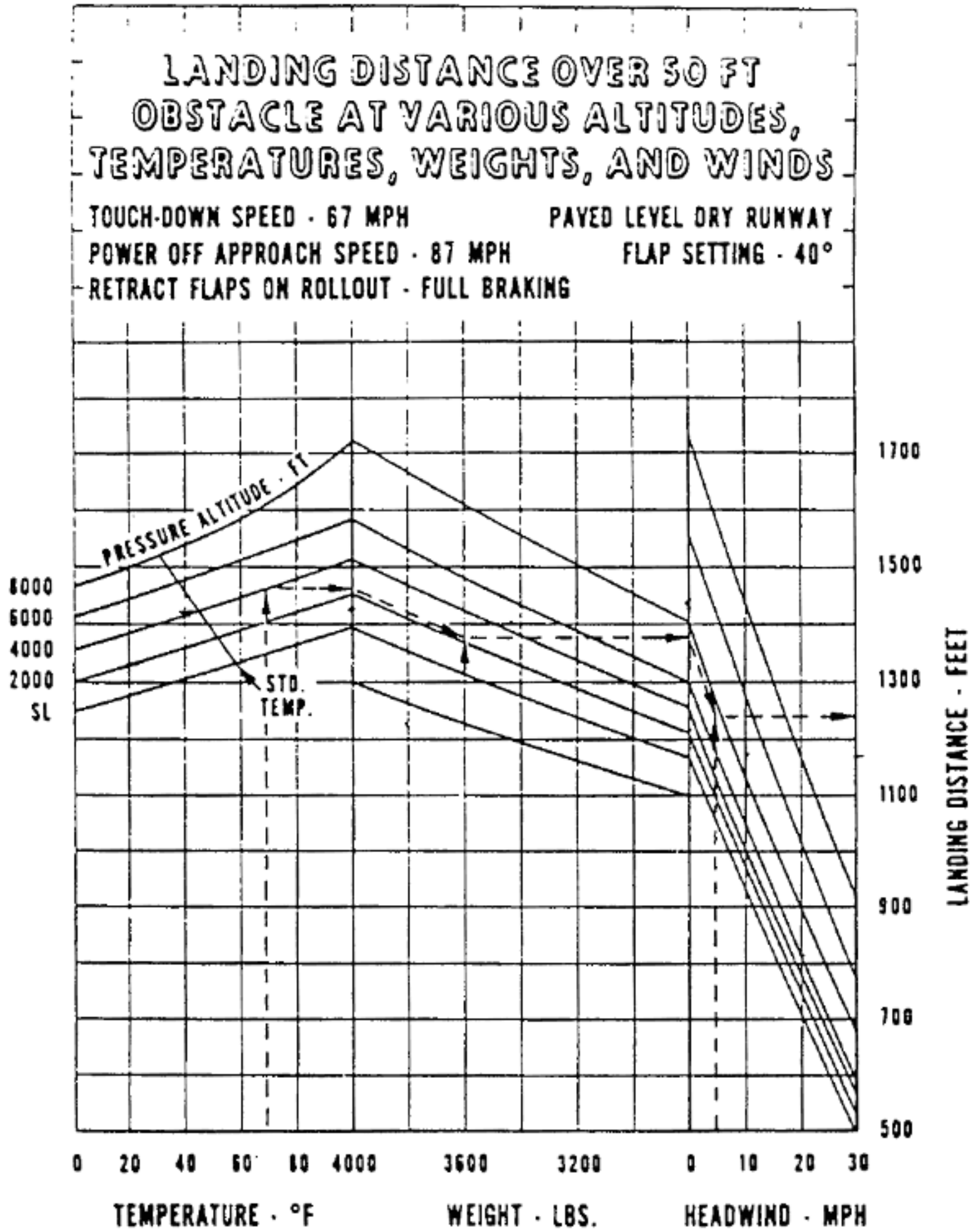
761 577



The above distances may be reduced by approximately 25% when the aircraft is equipped with optional Heavy Duty Wheels, Brakes and Tires. (Reference Aircraft Equipment List in Weight and Balance section of this manual.)

Example: Temp. 70° F Wt. 3600 lbs Ground Run 615 ft
 Press. Alt. 2000 ft Hd. wind 5 MPH

761577



The above distances may be reduced by approximately 12% when the aircraft is equipped with optional Heavy Duty Wheels, Brakes and Tires. (Reference Aircraft Equipment List in Weight and Balance section of this manual.)

Example: Temp. 70° F Wt. 3600 lbs. Landing Dist. 1240 ft
 Press. Alt. 4000 ft Hd. wind 5 MPH

761 577

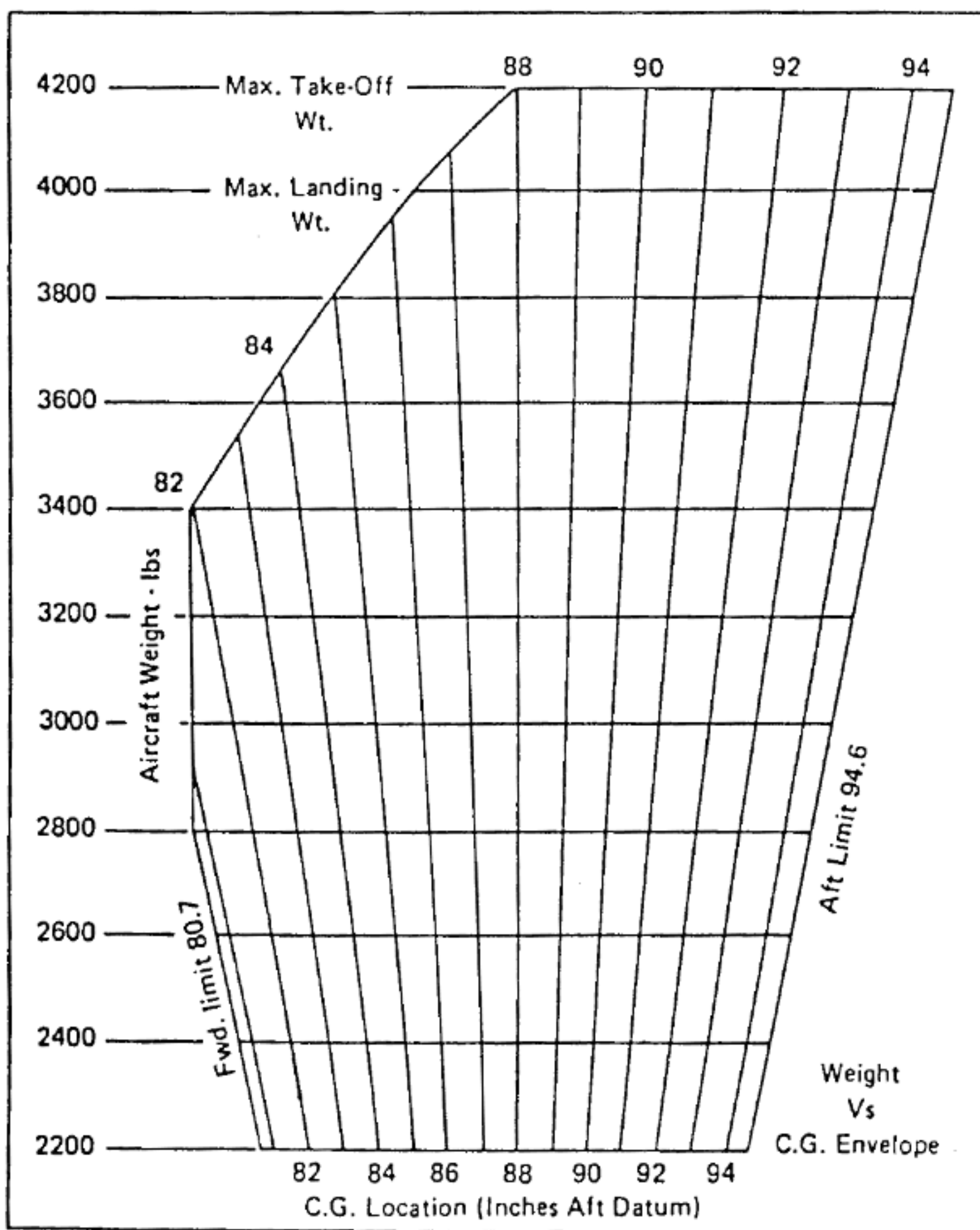
Power Setting Table - Lycoming Model IO-360-C Series, 200 HP Engine

Press. Alt Feet	Std. Alt Temp °F	110 HP - 55% Rated		130 HP - 65% Rated		150 HP - 75% Rated		Press. Alt Feet					
		RPM AND MAN. PRESS. 2100 2200 2300 2400	RPM AND MAN. PRESS. 2100 2200 2300 2400	RPM AND MAN. PRESS. 2100 2200 2300 2400	RPM AND MAN. PRESS. 2300 2400								
SL	59	22.9	22.0	21.0	20.4	25.9	24.8	23.8	22.9	26.5	25.5	25.5	SL
1,000	55	22.7	21.8	20.8	20.2	25.6	24.5	23.5	22.7	26.2	25.2	25.2	1,000
2,000	52	22.4	21.5	20.6	20.0	25.4	24.3	23.3	22.5	25.9	25.0	25.0	2,000
3,000	48	22.2	21.3	20.4	19.8	25.1	24.0	23.0	22.2	25.7	24.7	24.7	3,000
4,000	45	21.9	21.1	20.2	19.5	24.8	23.8	22.8	22.0	FT	24.4	24.4	4,000
5,000	41	21.7	20.8	20.0	19.3	FT	23.6	22.6	21.7	-	FT	FT	5,000
6,000	38	21.4	20.6	19.8	19.1	-	FT	22.3	21.5	-	22.3	21.5	6,000
7,000	34	21.2	20.4	19.6	18.9	-	-	22.1	21.3	-	22.1	21.3	7,000
8,000	31	21.0	20.1	19.4	18.7	-	-	FT	21.0	-	FT	21.0	8,000
9,000	27	FT	19.9	19.2	18.5	-	-	-	FT	-	-	FT	9,000
10,000	23	-	19.7	19.0	18.3	-	-	-	-	-	-	-	10,000
11,000	19	-	FT	18.7	18.1	-	-	-	-	-	-	-	11,000
12,000	16	-	-	FT	17.8	-	-	-	-	-	-	-	12,000
13,000	12	-	-	-	17.6	-	-	-	-	-	-	-	13,000
14,000	9	-	-	-	FT	-	-	-	-	-	-	-	14,000

To maintain constant power, correct manifold pressure approximately 0.16" Hg for each 10° F variation in inlet air temperature from standard altitude temperature. Add manifold pressure for air temperatures above standard; subtract for temperatures below standard.

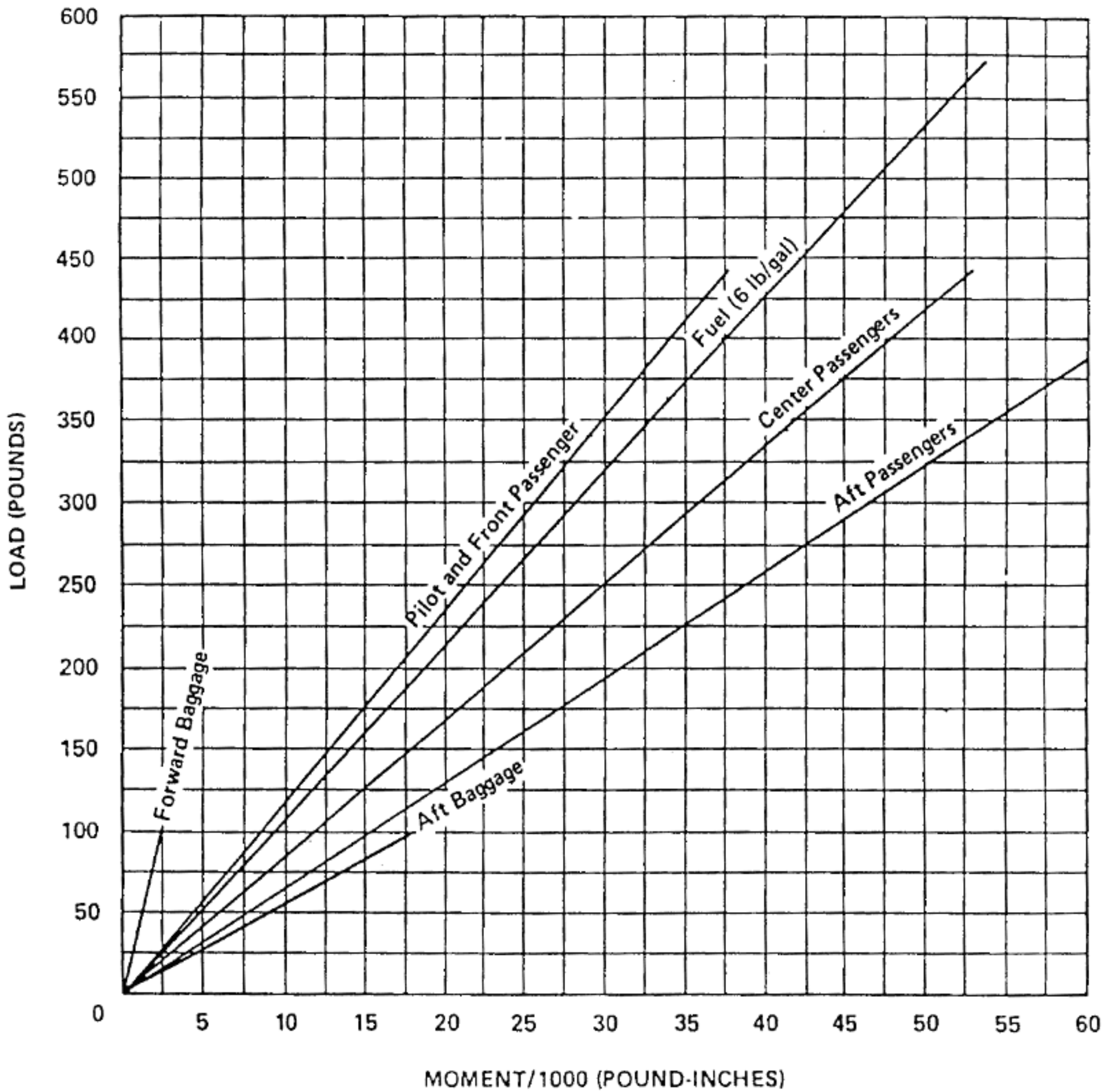
230 061, 731011

IT IS THE RESPONSIBILITY OF THE OWNER AND PILOT TO ASCERTAIN THAT THE AIRPLANE ALWAYS REMAINS WITHIN THE ALLOWABLE WEIGHT VS. CENTER OF GRAVITY ENVELOPE WHILE IN FLIGHT.

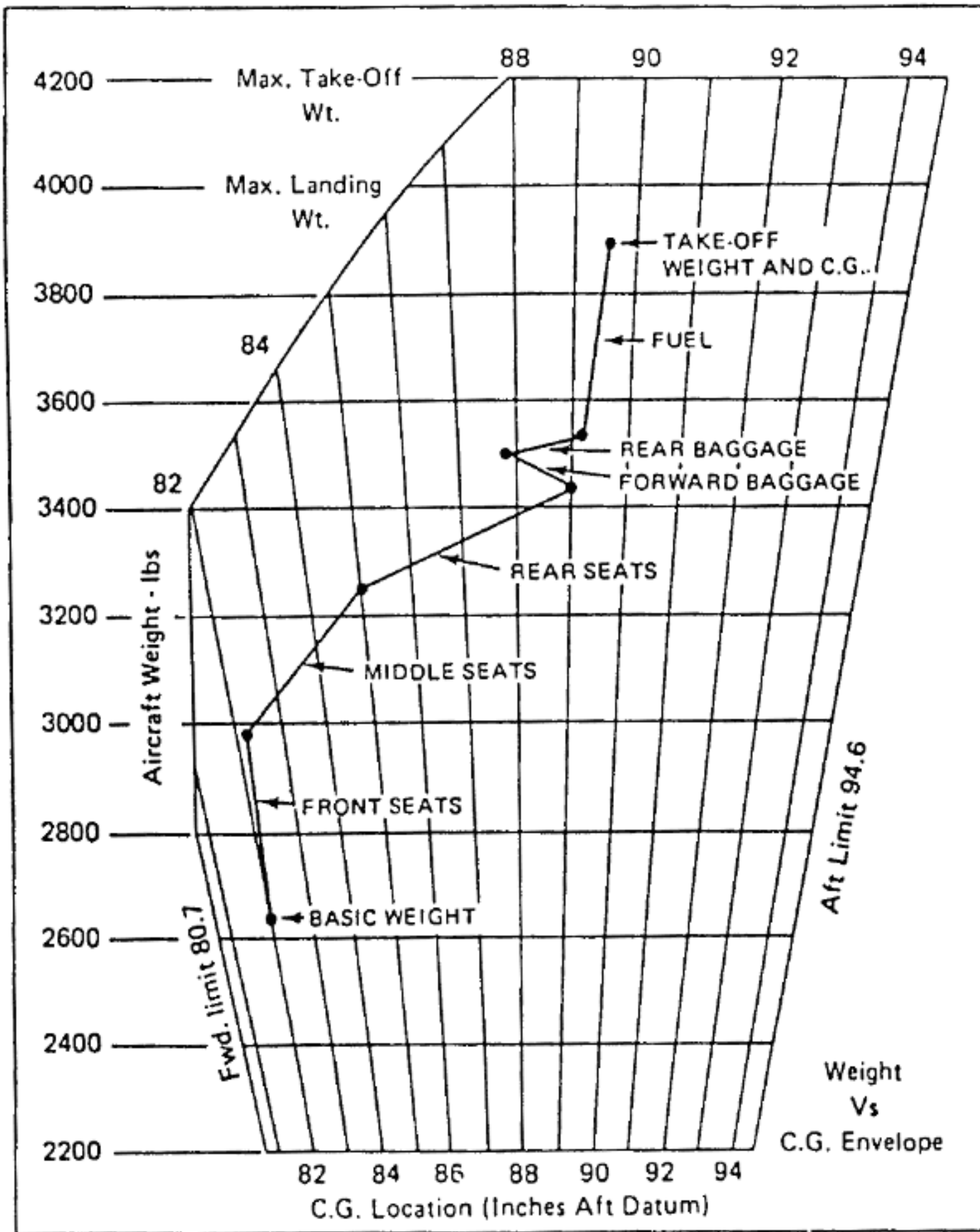


Moment change due to retracting Landing Gear = - 32 in. -lbs.

LOADING GRAPH



SAMPLE PROBLEM



Moment change due to retracting Landing Gear = -32 in.-lbs.